

Errata

Fuselage Boundary-Layer Effects on Sound Propagation and Scattering

H. Y. Lu

Boeing Commercial Airplanes,
Seattle, Washington 98124

[AIAAJ Vol. 28, No. 7, pp. 1180–1186 (1990)]

DUE to an error during the production of this paper, Eq. (3) was published incorrectly. The correct Eq. (3) is

$$p\omega = \sum_{n=-\infty}^{\infty} e^{in\theta} \int_{-\infty}^{\infty} F(r;n,k_x,\omega) e^{ik_x x} dk_x$$

Navier-Stokes Solutions for Vortical Flows over a Tangent-Ogive Cylinder

Peter-M. Hartwich

Vigyan, Inc., Hampton, Virginia
and

R. M. Hall

NASA Langley Research Center, Hampton, Virginia

[AIAAJ Vol. 28, No. 7, pp. 1171–1179 (1990)]

THE following revised figures should replace those published with this article:

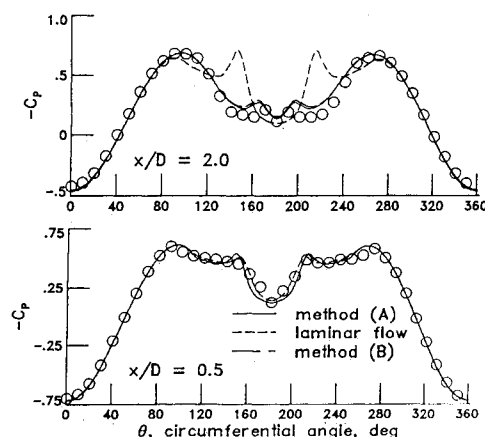
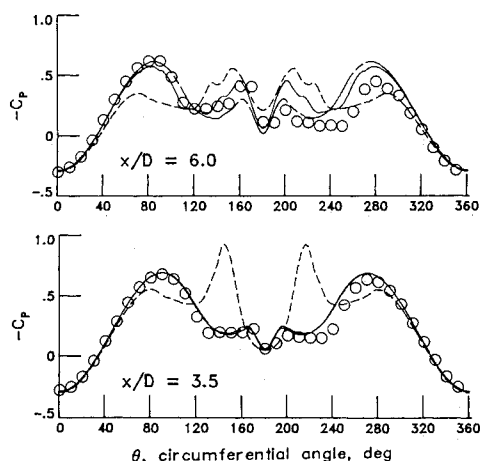


Fig. 6 Experimental (symbols) and computational (lines) circumferential surface pressure distributions: $\alpha = 20$ deg, $Re_D = 0.2$ million, $40 \times 73 \times 65$ grid points.

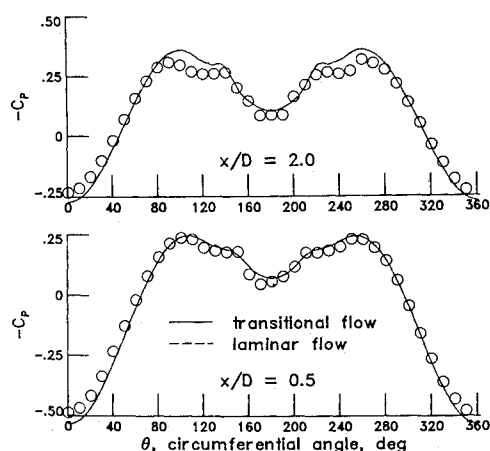
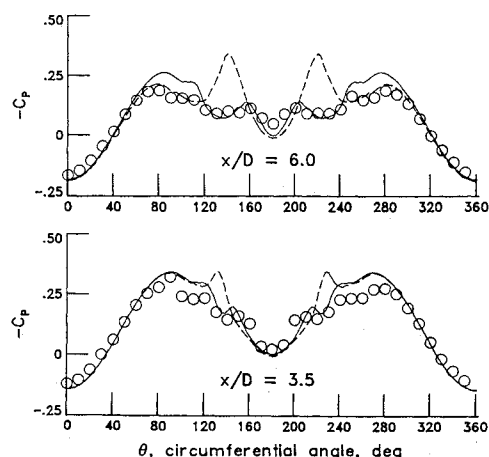


Fig. 8 Measured (symbols) and computed (lines) circumferential surface pressure distributions: $\alpha = 30$ deg, $Re_D = 0.8$ million, $40 \times 73 \times 65$ grid points.

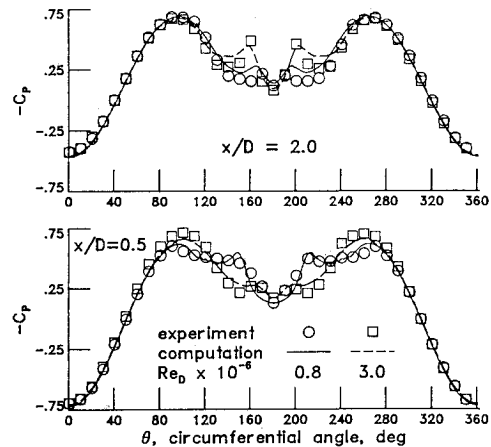
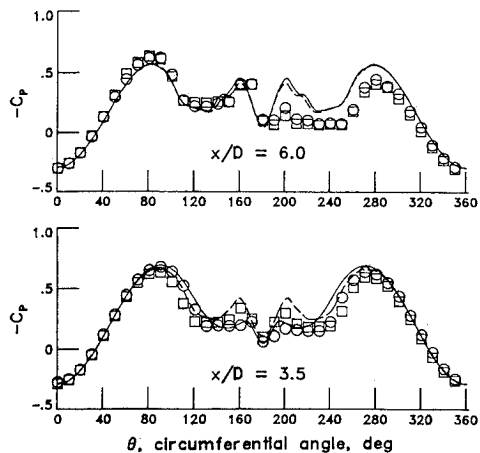


Fig. 14 Measured and computed circumferential pressure distributions: $\alpha = 30$ deg, $40 \times 91 \times 97$ grid points.